REMARKS

Entry and consideration of this Amendment are respectfully requested.

Respectfully submitted,

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APPENDIX

VERSION WITH MARKINGS TO SHOW CHANGES MADE

IN THE CLAIMS:

The claims are amended as follows:

2. (Four times amended) A rocket engine [nozzle] as claimed in claim 14, wherein the separation triggering elements comprise:

injection orifices positioned for injecting fluid through a wall of the nozzle body; and

at least two independent injection orifices being distributed over the perimeter of the wall of the nozzle body, each of the injection orifices constituting a discrete separation triggering element that induces a distinct zone of jet separation.

3. (Three Times Amended) A rocket engine [nozzle] as claimed in claim 2, wherein the injection orifices are uniformly distributed over the perimeter of the wall of the nozzle body.

4. (Four times amended) A rocket engine [nozzle] as claimed in claim 14, wherein the injection orifices comprise at least two, which are symmetrically positioned around the circumference of said divergent nozzle body.

5. (Three Times Amended) A rocket engine [nozzle] as claimed in claim 3, wherein the injection orifices comprise three in number and are arranged at substantially 120° to one another over the perimeter of the nozzle body.

6. (Three Times Amended) A rocket engine [nozzle] as claimed in claim 2, wherein said injection cross section is arranged at distance D from the throat which is substantially less than a distance D_0 of a location of spontaneous separation of the flow at sea level.

7. (Four times amended) A rocket engine [nozzle] as claimed in claim 6, said means for simultaneously injecting comprising:

a plurality of injectors situated at different distances from the throat for simultaneously injecting said fluid; and

a distributing device for selectively feeding said injectors at different cross sectional locations to take into account the variation of said distance of spontaneous separation of the flow as a function of altitude.

14. (Twice amended) A rocket engine [nozzle] comprising:

a combustion chamber;

a throat; and

a divergent nozzle body downstream of said throat, said nozzle body having an axis and a control system for controlling jet separation of a flow in the nozzle body, said [flow]thrust being parallel [to]with the axis of the nozzle body,

wherein said control system comprises,

[at least two]a plurality of mutually spaced separation triggering elements positioned on [at least one]an injection cross section of the divergent nozzle body [that is] perpendicular to the [nozzle] axis of the nozzle body, and

a means for simultaneously injecting fluid through the [at least two]mutually spaced separation triggering elements of said [at least one] injection cross section of the divergent nozzle body, forming a three-dimensional separation of said flow, [wherein said spacing of the separation triggering elements is sufficient for said injection through

the at least two separation triggering elements to generate as many] and for generating distinct zones of jet separation [as there are]corresponding to the spaced separation triggering elements from a respective plurality of mutually spaced initiation points positioned in the divergent nozzle body, [to form a three-dimensional separation of the flow]wherein said separation triggering elements are spaced so that said injection occurs through the separation triggering elements.

15. (Once amended) The rocket engine [nozzle] as claimed in claim 14, wherein the nozzle body is conical.

Claims 16-18 are added as new claims.